NATIONAL ADVISORY COMMITTEE
FOR AERONAUTICS

REPORT No. 885

FLIGHT INVESTIGATION ON A FIGHTER-TYPE AIRPLANE
OF FACTORS WHICH AFFECT THE LOADS AND
LOAD DISTRIBUTIONS ON THE VERTICAL TAIL
SURFACES DURING RUDDER KICKS AND
FISHTAILS

By JOHN BOSHAR

1947
AERONAUTIC SYMBOLS

1. FUNDAMENTAL AND DERIVED UNITS

<table>
<thead>
<tr>
<th>Unit</th>
<th>Symbol</th>
<th>Metric</th>
<th>English</th>
</tr>
</thead>
<tbody>
<tr>
<td>Length</td>
<td>l</td>
<td>meter</td>
<td>foot (or mile)</td>
</tr>
<tr>
<td>Time</td>
<td>l</td>
<td>second</td>
<td>second (or hour)</td>
</tr>
<tr>
<td>Force</td>
<td>F</td>
<td>weight of 1 kilogram</td>
<td>weight of 1 pound</td>
</tr>
<tr>
<td>Power</td>
<td>P</td>
<td>horsepower (metric)</td>
<td>horsepower</td>
</tr>
<tr>
<td>Speed</td>
<td>V</td>
<td>kilometers per hour</td>
<td>miles per hour</td>
</tr>
<tr>
<td></td>
<td></td>
<td>meters per second</td>
<td>feet per second</td>
</tr>
</tbody>
</table>

2. GENERAL SYMBOLS

- \( W \): Weight = \( mg \)
- \( g \): Standard acceleration of gravity = 9.80665 m/s² or 32.1740 ft/sec²
- \( m \): Mass
- \( I \): Moment of inertia = \( ml^2 \). (Indicate axis of radius of gyration \( k \) by proper subscript.)
- \( \mu \): Coefficient of viscosity
- \( \nu \): Kinematic viscosity
- \( \rho \): Density (mass per unit volume)
- \( \Omega \): Resultant angular velocity
- \( \alpha \): Angle of attack
- \( \beta \): Angle of downwash
- \( \alpha_i \): Angle of stabilizer setting (relative to thrust line)
- \( \alpha_r \): Resultant moment
- \( \alpha_i \): Angle of attack, infinite aspect ratio
- \( \alpha_a \): Angle of attack, induced
- \( \gamma \): Flight-path angle
- \( V \): True air speed
- \( q \): Dynamic pressure, \( \frac{1}{2} \rho V^2 \)
- \( L \): Lift, absolute coefficient \( C_l = \frac{L}{qS} \)
- \( D \): Drag, absolute coefficient \( C_d = \frac{D}{qS} \)
- \( D_p \): Profile drag, absolute coefficient \( C_{D_p} = \frac{D_p}{qS} \)
- \( D_i \): Induced drag, absolute coefficient \( C_{D_i} = \frac{D_i}{qS} \)
- \( D_s \): Parasite drag, absolute coefficient \( C_{D_s} = \frac{D_s}{qS} \)
- \( G \): Cross-wind force, absolute coefficient \( C_G = \frac{G}{qS} \)

3. AERODYNAMIC SYMBOLS

- \( B \): Reynolds number, \( \frac{\rho V l}{\mu} \) where \( l \) is a linear dimension (e.g., for an airfoil of 1.0 ft chord, 100 mph, standard pressure at 15° C, the corresponding Reynolds number is 935,400; or for an airfoil of 1.0 m chord, 100 mps, the corresponding Reynolds number is 8,865,000)
REPORT No. 885

FLIGHT INVESTIGATION ON A FIGHTER-TYPE AIRPLANE
OF FACTORS WHICH AFFECT THE LOADS AND
LOAD DISTRIBUTIONS ON THE VERTICAL TAIL
SURFACES DURING RUDDER KICKS AND
FISHTAILS

By JOHN BOSHAR

Langley Memorial Aeronautical Laboratory
Langley Field, Va.
National Advisory Committee for Aeronautics

Headquarters, 1724 F Street NW, Washington 25, D. C.

Created by act of Congress approved March 3, 1915, for the supervision and direction of the scientific study of the problems of flight (U. S. Code, title 49, sec. 241). Its membership was increased to 15 by act approved March 2, 1929. The members are appointed by the President, and serve as such without compensation.

Jerome C. Hunsaker, Sc. D., Cambridge, Mass., Chairman

Alexander Wetmore, Sc. D., Secretary, Smithsonian Institution, Vice Chairman

Hon. John R. Alison, Assistant Secretary of Commerce.

Vannevar Bush, Sc. D., Chairman, Research and Development Board, Department of National Defense.

Edward U. Condon, Ph. D., Director, National Bureau of Standards.

Donald B. Duncan, Vice Admiral, Deputy Chief of Naval Operations (Air).

R. M. Hazen, B. S., Chief Engineer, Allison Division, General Motors Corp.

William Littlewood, M. E., Vice President, Engineering, American Airlines System.

Theodore C. Lonquest, Rear Admiral, Assistant Chief for Research and Development, Bureau of Aeronautics, Navy Department.

Hugh L. Dryden, Ph. D., Director of Aeronautical Research

John W. Crowley, Jr., B. S., Associate Director of Aeronautical Research

Edward M. Powers, Major General, United States Air Force, Deputy Chief of Staff, Materiel.

Arthur E. Raymond, M. S., Vice President, Engineering, Douglas Aircraft Co.

Francis W. Reichelderfer, Sc. D., Chief, United States Weather Bureau.

Carl Spaatz, General, Chief of Staff, United States Air Force.

Orville Wright, Sc. D., Dayton, Ohio.

Theodore P. White, Sc. D., Administrator of Civil Aeronautics, Department of Commerce.

TECHNICAL COMMITTEES

AERODYNAMICS

Power Plants for Aircraft

Aircraft Construction

Operating Problems

Self-Propelled Guided Missiles

Industry Consulting

Coordination of Research Needs of Military and Civil Aviation

Preparation of Research Programs

Allocation of Problems

Prevention of Duplication

Consideration of Inventions

Langley Memorial Aeronautical Laboratory, Langley Field, Va.

Ames Aeronautical Laboratory, Moffett Field, Calif.

Flight Propulsion Research Laboratory, Cleveland Airport, Cleveland, Ohio

Conduct, under unified control, for all agencies, of scientific research on the fundamental problems of flight

Office of Aeronautical Intelligence, Washington, D. C.

Collection, classification, compilation, and dissemination of scientific and technical information on aeronautics
REPORT No. 885

FLIGHT INVESTIGATION ON A FIGHTER-TYPE AIRPLANE OF FACTORS WHICH AFFECT THE LOADS AND LOAD DISTRIBUTIONS ON THE VERTICAL TAIL SURFACES DURING RUDDER KICKS AND FISHTAILS

By John Boshar

SUMMARY

Results are presented of a flight investigation conducted on a fighter-type airplane to determine the factors which affect the loads and load distributions on the vertical tail surfaces in maneuvers. An analysis is made of the data obtained in steady flight, rudder kicks, and fishtail maneuvers.

For the rudder kicks, the significant loads were the "deflection load" resulting from an abrupt control deflection and the "dynamic load" consisting of a load corresponding to the new static equilibrium condition for the rudder deflected plus a load due to a transient overshoot. The deflection load is proportional to the angular acceleration which in turn is dependent upon the rate and amount of control deflection and upon the directional response characteristics of the airplane. The dynamic load had an angular acceleration load superposed on it as a result of the rudder being reversed at the time of maximum sideslip. The critical loads on the rudder were associated with the deflection load, and those on the fin, with the dynamic load.

The minimum time to reach the maximum control deflection attainable by the pilot in any flight condition was found to be a constant.

In the fishtail maneuvers, it was found that the pilot tended to deflect the rudder in phase with the natural frequency of the airplane. At the condition of resonance the load on the fin and that on the rudder are approximately 90° out of phase. The maximum loads measured in fishtails were of the same order of magnitude as those from a rudder kick in which the rudder deflection angle is maximum.

INTRODUCTION

The problem of evolving methods for designing the tail surfaces of fighter-type airplanes for the dynamic effects which occur in maneuvers has received much attention in recent years. In the case of the horizontal tail, methods by which the loads may be determined for an arbitrary type of elevator motion have been introduced (references 1 and 2) and the type of control deflection to be assumed in design has been specified (reference 3).

In the case of the vertical tail, however, the current design specifications consider only steady-state conditions for loads associated with a specified steady yaw or a specified rudder angle. Indications have been that the loads on the vertical tail are more critical in maneuvers than in steady-flight conditions. For instance, in reference 4, critical vertical-tail loads in rolling pull-out maneuvers were shown to be related to the ratio of aileron power and the static directional-stability derivative of the airplane; whereas, in reference 5 the dynamic loads in abrupt rudder kicks or in fishtail maneuvers were shown to reach high values. For some time, therefore, there has existed a need for a systematic flight investigation to evaluate the factors which influence the vertical-tail loads.

The purpose of the present paper is to present the results of a flight investigation of the factors which affect the loads and the load distributions on the vertical tail surfaces in rudder kicks and fishtail maneuvers. An attempt has been made to isolate the effects of power, of speed, of initial sideslip, and of rate, amount, and direction of control deflection. Emphasis has been placed upon the presentation of the experimental results in the light of theoretical considerations.

SYMBOLS

- \( \delta_r \) rudder deflection angle, degrees
- \( \dot{\delta}_r \) maximum rate of rudder deflection, degrees per second
- \( \delta_e \) elevator deflection angle, degrees
- \( \beta \) sideslip angle, degrees
- \( F \) pedal force, pounds
- \( N_v \) normal force on vertical tail, pounds
- \( N_r \) normal force on rudder, pounds
- \( N_f \) normal force on fin, pounds
- \( N_{r1} \) first load peak on vertical tail, pounds
- \( N_{r2} \) second load peak on vertical tail, pounds
- \( N_{r3} \) second load peak on rudder, pounds
- \( N_{f1} \) first load peak on fin, pounds
- \( N_{f2} \) second load peak on fin, pounds
- \( C_{nv} \) normal-force coefficient on vertical tail \( (N_v/qS_v) \)
- \( C_{nr} \) normal-force coefficient on rudder \( (N_r/qS_r) \)
- \( C_{nf} \) normal-force coefficient on fin \( (N_f/qS_f) \)

With the foregoing symbols, the prefix \( \Delta \) represents an increment; for maneuvers, it indicates the maximum increment measured from the initial steady-flight value; for steady sideslip, it represents an increment measured from the trim value for wings level.
V

airspeed, miles per hour

\( V_e \)
equivalent airspeed, miles per hour (\( V e^{1/2} \))

\( S_t \)
total vertical tail area, square feet

\( x_r \)
distance from center of gravity to rudder hinge line (absolute value), feet

\( q \)
dynamic pressure, pounds per square foot (\( \frac{1}{2} \rho V^2 \))

\( I_z \)
moment of inertia about Z-axis, pound-foot-second^2

\( T \)
propeller thrust, pounds

\( Q_e \)
torque coefficient (\( Q / \rho V^2 D^3 \))

\( Q \)
propeller torque, pound-feet

\( D \)
propeller diameter, feet

\( b \)
wing span, feet

\( S \)
wing area, square feet

\( P \)
pressure coefficient (\( (p-p_0)/q \))

\( p \)
local static pressure

\( p_0 \)
free-stream static pressure

\( \rho \)
mass density of air, slugs per cubic foot

\( \rho_0 \)
mass density of air at sea level, slugs per cubic foot

\( C_n \)
yawing-moment coefficient, tail off (\( N'/qSb \))

\( \sigma = \frac{p}{p_0} \)
maximum yawing velocity, radians per second

\( \dot{\psi} \)
angular acceleration in yaw, radians per second

\( \dot{\psi}_1 \)
first maximum angular acceleration in yaw, radians per second

\( \dot{\psi}_2 \)
second maximum angular acceleration in yaw, radians per second

\( \dot{\delta} \)
maximum pitching velocity, radians per second

\( \dot{\delta}_1 \)
first maximum angular acceleration in pitch, radians per second

\( \Delta t \)
time interval during which maneuver is allowed to continue before rudder is returned to zero, seconds

\( \Delta \alpha \)
increment in angle of attack of vertical tail, degrees

\( \frac{dC_n}{d\delta} \)
rate of change of yawing-moment coefficient with sideslip angle (tail off)

\( \frac{dC_n}{d\beta} \)
measured rate of change of normal-force coefficient on vertical tail with angle of sideslip, including the effect of rudder deflection

\( \frac{d\beta}{d\delta_1} \)
rate of change of sideslip with change in rudder angle (from steady sideslip measurements)

\( \frac{dC_l}{d\delta_1} \)
estimated rate of change of lift coefficient with control deflection for isolated vertical tail (1.10 per radian)

\( \frac{dC_l}{d\alpha} \)
estimated rate of change of lift coefficient with angle of attack for isolated vertical tail (1.43 per radian)

\( \frac{d\alpha}{d\delta_1} \)
estimated rudder effectiveness (0.77)

DEFINITIONS

Deflection load: Maximum increment in load due to abrupt control deflection at the start of maneuver (first load peak).

Dynamic load: Maximum increment in load including load due to the static balance condition for rudder deflected, load due to transient overshoot, and load due to rudder reversal (second load peak).

U-type control manipulation: Hypothetical control manipulation in which both the initial kick and the return of rudder have the same amount and rate of control deflection.

APPARATUS

Test airplane.—The investigation was conducted on a modified Curtiss P-40K airplane which is a low-wing fighter airplane with a gross weight of about 8200 pounds and equipped with a V-1710-F4R Allison engine rated at 1000 horsepower at a pressure altitude of 10,800 feet. Figure 1 shows photographs of the test airplane. Figure 2 presents a three-view drawing of the airplane; table I contains a list of some pertinent geometric characteristics.

The military equipment, radio, and fuselage gas tanks were removed to permit the installation of the recording instruments. The airplane was flown with a center-of-gravity location of 29.5 percent of the mean aerodynamic chord.

Tail surfaces.—In order to improve the directional stability characteristics and to permit the pilot to fly more easily...
through the speed range with only one setting of the rudder-trim tab, a fin extension was added (see fig. 3), and the fin offset was changed from $1\frac{1}{2}^\circ$ left to $0^\circ$ offset as suggested in reference 6.

The horizontal tail surfaces were unchanged with the exception of the fairing added at the juncture of the fin and horizontal tail to cover the pressure lines. The amount of protuberance of this fairing is shown in the photographs in figure 4.

Orifices were installed opposite each other on the left and right sides of the vertical tail at the locations shown in figure 5.

**Flight instruments.**—Instruments installed to measure the differential pressures, the control forces, the control deflections, and the motions of the airplane were as follows:

1. Multicell manometers to measure the differential pressures over the vertical tail surface at the points shown in figure 5.
2. An NACA airspeed recorder with the swivelling static head located approximately one chord forward of the right wing tip. (See fig. 1 (a).)
3. Control-force recorders which measured the forces exerted by the pilot on the stick (aileron and elevator) and on the rudder pedals.
4. NACA electrical control-position recorders which measured the elevator- and rudder-control positions at points on these controls near the fuselage center line.
5. A sideslip-angle recorder mounted approximately one-half chord above and one chord forward of the left wing tip. (See fig. 1 (a).)
6. Accelerometers which recorded transverse and normal accelerations at points 59 and 152 inches behind the center of gravity.
7. Turnmeters which measured the angular velocities in yaw, pitch, and roll.
8. A timer used to synchronize all records.

Prior to each test the pilot noted the manifold pressure, the pressure altitude, the airspeed, and the cockpit settings of the rudder, elevator, and aileron trim tabs.

**TEST PROGRAM**

The test program may be divided into three parts: (1) tests conducted to obtain steady-flight data, (2) tests in which rudder kicks were made, and (3) tests in which fishtail maneuvers were made. All speeds mentioned are equivalent airspeeds.

**Steady-flight runs.**—Inasmuch as the vertical-tail loads on an airplane are related to its steady-sideslip characteristics, a number of steady-flight runs were made at various values of steady sideslip and speed, and at two power conditions. The data were recorded after the pilot had trimmed the airplane at the test condition. Runs were obtained through a speed range of 100 to 380 miles per hour with power on (power for level flight or rated power when necessary) and 100 to 220 miles per hour with power off.

**Rudder kicks.**—Rudder kicks (single abrupt rudder deflections) are useful in the study of the directional stability characteristics of an airplane and for the investigation of the effects of rate, amount, and direction of control deflection on the vertical-tail loads.

A total of approximately 50 left and right rudder kicks were made during which pressure distributions were measured. Of these runs, approximately 30 were kicks from the wings-level condition and 20 were kicks against
FIGURE 3.—Plan form of vertical tail used on test airplane and profiles of the airfoil sections around which pressure orifices were distributed.

FIGURE 4.—Vertical tail showing profile and plan form of protuberance caused by fairing over pressure lines.

an initial steady sideslip. The runs were made at speeds of approximately 100, 200, and 300 miles per hour with power on and power off. The rudder kicks were performed at medium and fast rates from trimmed flight. In addition, 70 rudder kicks in which loads were not measured were found to be useful in the analysis.

Fishtail maneuvers.—Fishtail maneuvers (periodic rudder oscillations) were made with power off and power on at speeds of 150 and 200 miles per hour during which the pilot attempted to maximize the loads on the vertical tail. Also, runs were made at 150 miles per hour during which the pilot applied an abrupt rudder deflection against the swing at the time of maximum yawing velocity. A second pilot was asked to perform mild fishtail maneuvers at speeds of 200, 250, 300, and 350 miles per hour. For this series the pilot was free to use as much coordination as he wished so that information would be obtained to evaluate the maneuver under such conditions.
LOADS AND LOAD DISTRIBUTIONS ON THE VERTICAL TAIL SURFACES DURING RUDDER KICKS AND FISHTAILS

<table>
<thead>
<tr>
<th>Rib</th>
<th>Chord (in.)</th>
<th>1</th>
<th>2</th>
<th>3</th>
<th>4</th>
<th>5</th>
<th>6</th>
<th>7</th>
<th>8</th>
<th>9</th>
<th>10</th>
<th>11</th>
<th>12</th>
</tr>
</thead>
<tbody>
<tr>
<td>T</td>
<td>34.0</td>
<td>11.3</td>
<td>23.1</td>
<td>29.4</td>
<td>52.6</td>
<td>65.9</td>
<td>70.2</td>
<td>89.4</td>
<td>90.4</td>
<td>93.4</td>
<td>81.1</td>
<td>96.7</td>
<td></td>
</tr>
<tr>
<td>U</td>
<td>42.2</td>
<td>4.1</td>
<td>10.3</td>
<td>23.4</td>
<td>55.4</td>
<td>62.5</td>
<td>59.0</td>
<td>72.2</td>
<td>60.7</td>
<td>62.1</td>
<td>60.4</td>
<td>60.7</td>
<td></td>
</tr>
<tr>
<td>V</td>
<td>52.4</td>
<td>3.3</td>
<td>13.3</td>
<td>26.4</td>
<td>57.8</td>
<td>44.8</td>
<td>53.8</td>
<td>57.0</td>
<td>58.7</td>
<td>62.1</td>
<td>60.4</td>
<td>60.7</td>
<td></td>
</tr>
<tr>
<td>W</td>
<td>56.0</td>
<td>3.1</td>
<td>6.0</td>
<td>16.4</td>
<td>18.3</td>
<td>27.8</td>
<td>41.1</td>
<td>51.7</td>
<td>56.4</td>
<td>59.6</td>
<td>60.7</td>
<td></td>
<td></td>
</tr>
<tr>
<td>X</td>
<td>26.6</td>
<td>7.4</td>
<td>37.1</td>
<td>65.3</td>
<td>66.9</td>
<td>53.6</td>
<td>53.7</td>
<td>52.8</td>
<td>57.4</td>
<td>59.1</td>
<td>53.6</td>
<td></td>
<td></td>
</tr>
<tr>
<td>Y</td>
<td>31.2</td>
<td>10.9</td>
<td>41.4</td>
<td>71.8</td>
<td>89.4</td>
<td>57.3</td>
<td>57.3</td>
<td>57.3</td>
<td>57.3</td>
<td>57.3</td>
<td>57.3</td>
<td>57.3</td>
<td></td>
</tr>
<tr>
<td>Z</td>
<td>22.4</td>
<td>7.5</td>
<td>40.4</td>
<td>75.6</td>
<td>75.6</td>
<td>75.6</td>
<td>75.6</td>
<td>75.6</td>
<td>75.6</td>
<td>75.6</td>
<td>75.6</td>
<td>75.6</td>
<td></td>
</tr>
</tbody>
</table>

**METHODS**

**Pressure distributions.**—The records used in evaluating the pressure distributions were read at time values which would permit an accurate time history to be represented. The chordwise integrations were performed in two parts so that the chordwise and spanwise loads could be obtained separately for the fin and rudder. A numerical method of obtaining the spanwise center of load on the fin was used.

**Other records.**—The angle of sideslip for the steady-sideslip results was corrected for the effect of inflow as determined from the results of a calibration flight in which similar sideslip-angle recorders were installed on each wing tip. This correction was not made for the sideslip-angle records in the time histories since only incremental values were used in the analysis and the angle of inflow correction was nearly constant throughout the maneuver.

The only other corrections made were the compressibility correction to the airspeed and the correction to the rudder and elevator angles for the amount of trim-tab deflection required to keep the wings in level trim.

The rate of control deflection and angular accelerations were obtained by mechanically differentiating the control deflection and the angular-velocity records, respectively.

**Separation of load components.**—The method of separation of load components on the vertical tail was found to be accomplished most conveniently by considering the load to be made up of two components: one necessary to balance the unstable wing-fuselage yawing moment in sideslip and one due to yawing acceleration, or

\[ \Delta N_i = \Delta \beta \frac{dC_n}{d\beta} \frac{b}{x_e} q S_e - I_{y_0} \]

(1)

However, some use was also made of the expression for the load in terms of effective angle of attack at the tail; that is,

\[ \Delta N_i = \Delta \alpha_e \left( \frac{dC_{\alpha}}{d\alpha} \right)_e q S_e \]

(2)

where, approximately,

\[ \Delta \alpha_e = -\beta + \left( \frac{d\alpha}{d\delta_e} \right)_e \Delta \delta_e \]

The form of equation (1) is particularly useful in the present case because both the parameter \( dC_n/d\beta \) and the factor \( I_{y_0}/x_e \) were derivable from flight results as shown subsequently herein and also because the maximum loads could be defined when only the value of maximum yawing acceleration \( \dot{\psi} \) and the maximum angle of sideslip \( \Delta \beta \) were known.
RESULTS AND DISCUSSION—STEADY FLIGHT

Wings level.—The pertinent data obtained from tests with wings level are shown plotted in figures 6 and 7. Figure 6 shows the variation with speed of the amount of rudder, elevator, and sideslip angle required to maintain wings level for power on and power off. Figure 7 shows the variation of the normal-force coefficients over the fin, rudder, and total vertical tail, and the spanwise variation of center of load on the fin with speed. These curves are typical for a single-engine airplane. The variations shown in figures 6 and 7 are caused by the effects of propeller rotation in producing a twisting slipstream and by a direct asymmetric thrust due to the inclined propeller. With power off the variations are probably the result of a windmilling propeller, particularly at speeds lower than 200 miles per hour where the amount of blade adjustment possible is insufficient to maintain the rotation of the constant-speed propeller. The spanwise center of load on the fin moves outboard with decreasing speed but, from consideration of the loads, this movement with wings level is not very significant because of the small bending moments involved.

Steady sideslip.—Steady-sideslip data are presented in table II and in figures 8 to 12. The data are shown as incremental values measured from the condition with wings level.

Figure 8 presents the changes in rudder deflection, rudder pedal force, and elevator deflection required for changes in sideslip measured from the wings-level trim value. The increments in pedal force are shown as pedal-force factors, which are obtained by dividing the pedal force by the dynamic pressure so that the data from all speeds may be combined.

The change in elevator angle required with a change in sideslip results from a change in the pitching moment of the airplane with sideslip. The variation of rudder angle with angle of sideslip is seen to be approximately linear throughout the speed range. Figure 9 presents the variation of the normal-force coefficient with sideslip for the rudder, fin, and total vertical tail surface. The variations shown are consistent with the trends of figure 8. The rate of change of normal-force coefficient on the vertical tail with angle of sideslip \( \left( \frac{dC_n}{d\beta} \right) \), is used to define the load required on the vertical tail to balance the unstable yawing moment of the wing-fuselage configuration. From this value the parameter \( \frac{dC_n}{d\beta} \) may be obtained as

\[
\frac{dC_n}{d\beta} = \left( \frac{dC_n}{d\beta} \right)_{s} b S
\]

Figure 10 presents isometric views of the pressure distribution over the vertical tail at various incremental values of sideslip for power on at an airspeed of 220 miles per hour. The spanwise load distributions on the fin and rudder corresponding to the isometric diagrams of figure 10 are shown in figure 11.
LOADS AND LOAD DISTRIBUTIONS ON THE VERTICAL TAIL SURFACES DURING RUDDER KICKS AND FISHTAILS

Figures 8.-Variation of increments of rudder and elevator control deflections and pedal-force factor with incremental change in sideslip measured from wings in level flight with power on and power off.

Figure 12 shows the variation of spanwise center of load on the fin with change in sideslip from the wings-level trim value at airspeeds of 100, 160, and 220 miles per hour. With change in sideslip from the wings-level condition, according to figure 12, an inboard movement of the spanwise center of load occurs which is probably a result of the displacement of the tail from the region of greatest fuselage boundary layer.

RESULTS AND DISCUSSION—RUDDER KICKS

TIME HISTORIES

Data pertaining to the rudder kicks are plotted in figures 13 to 41. The data for all the rudder kicks are shown in tables III and IV. Before a detailed analysis of the loads is made, it would be of value to note the general nature of the airplane motion and the sequence or events. For this purpose typical time histories of the measurements are shown in figures 13 to 18.

Figures 13 and 15 present the time histories of right and left rudder kicks, respectively, made at airspeeds of 100, 200, and 300 miles per hour with power on. The normal load on the fin, rudder, and total vertical tail surfaces associated with these measurements are shown in figures 14 and 16. Time histories for two rudder kicks applied against initial steady sidslips to the left and right made at airspeeds of 200 miles per hour are shown in figure 17 and corresponding normal loads on the vertical tail surfaces, in figure 18.
From a study of the time histories the following sequence of events and items of interest may be observed:

1. Before the maneuver is started, the airplane is in steady trim flight as indicated by the constant initial values of the variables.
2. After the application of an abrupt pedal force a lag of the order of a fraction of a second occurs before the rudder begins to respond because of flexibility in the control system.
3. The airplane begins to yaw as soon as the rudder is deflected.
4. The greatest rate of change of yawing velocity (the maximum yawing acceleration) following the rudder deflection occurs before the value of sideslip has changed from the trim condition.
5. The time interval from the start of the maneuver to the time the maximum yawing velocity is reached is, roughly, inversely proportional to the airspeed.

The time histories show that an appreciable amount of pitching is induced during the maneuver. With right rudder deflection the pitching is nose-down and with left rudder deflection it is nose-up. The pitching is caused primarily by two effects; namely, the precessional moment which results from yawing the propeller disk and the change in airplane pitching moment with sideslip. The precessional effect
leads the effect of sideslip by a phase relation of approximately 90° since it depends upon the yawing velocity rather than the angle of yaw. Also, the sign of the precessional pitching moment depends upon the direction of yawing; whereas the sign of the airplane pitching moment due to sideslip is negative regardless of sideslip direction, as is shown by the variation of elevator required with sideslip (fig. 8). The net effects are additive for right rudder kicks and canceling for left rudder kicks. This result explains the phase difference between the yawing-velocity curve and the pitching-velocity curve for left and right rudder kicks. The combined effects for right rudder kicks produce a decrement in vertical acceleration as high as approximately 1.7g at the center of gravity, as is indicated by figure 13 (c).

The time histories of the loads on the vertical tail surfaces (figs. 14, 16, and 18) exhibit the same general characteristics as the load variation on the horizontal tail following an abrupt elevator deflection. The first significant feature is the load peak due to the abrupt deflection of the rudder. The second significant feature is the build-up of load in the opposite direction as the airplane responds to the unbalance created by the control deflection. In seeking to assume a new static equilibrium position a transient “overshoot” occurs, the magnitude of which is a function of the dynamic lateral stability of the airplane. The maximum balance load thus consists of a static-balance trim value and a transient load. This second load-peak increment is referred to as the “dynamic load.”

The load variation with time on the rudder and fin shows that the rudder carries most of the deflection load; whereas the fin carries most of the dynamic load.

The deflection load and dynamic load will be discussed separately, use being made of the breakdown of the load into the component necessary to balance the unstable yawing moment of the wing-fuselage combination and that associated with the yawing acceleration. (See section entitled “Methods.”) A time history of the component of load due to each factor and a comparison of the combined effects with the measured vertical-tail loads is shown in figure 19 for flight 11a, run 1. As expected, the agreement is particularly good since the parameter \( dC_{Ld}/d\Phi \) (already shown) and the factor \( I_{z}/x_{r} \) were determined with the aid of experimental results. The details of determining \( I_{z}/x_{r} \) will be given in the following section.

In the subsequent discussion the definitions illustrated in figure 20 may be helpful.

**DEFLECTION LOAD**

**General relations.** In the deflection load, as shown in figure 19, the component of load necessary to balance the unstable wing-fuselage moments in sideslip is absent and the deflection load is defined by the angular-acceleration component only; therefore, when the values of the first yawing acceleration \( \dot{\psi}_{1} \), the moment of inertia of the airplane \( I_{z} \), and the tail length \( x_{r} \) are known, the load may be determined by the relation

\[
\Delta N_{1} = \frac{I_{z}}{x_{r}} \dot{\psi}_{1}
\]

This relation is shown in figure 21 in which the maximum yawing acceleration \( \dot{\psi}_{1} \) is seen to be linearly related to the experimentally determined deflection load. This curve, then, is an experimental determination of the factor \( I_{z}/x_{r} \). Inasmuch as figure 21 shows that such a definite relationship exists, it will be used in the subsequent analysis to determine the deflection load from the value of yawing acceleration only. This relationship permits determination of tail loads by use of the rudder-kick data presented in table IV for which direct tail-load measurements were not available.

As an introduction to the factors which affect the magnitude of the deflection load, it is convenient to consider two extremes of control manipulation—zero and infinite rates of rudder deflection. When the rate of rudder deflection is zero or very slow, the airplane will adjust itself to a new static equilibrium position as each infinitesimal increment of unbalance is impressed and the deflection load will be zero.
regardless of the amount of control deflection or the airplane stability or mass characteristics. When the rate of rudder deflection is infinite, however, because of the inertia about the Z-axis, the lift is experienced before the airplane can respond and the deflection load becomes approximately equal to that on an isolated tail with a value corresponding to the amount of control deflection attained, that is,

$$\Delta N_{\text{f1}} = \left( \frac{dC_{\text{u}}}{d\delta_{\text{r}}} \right) \Delta \delta_{\text{r}} q S_{\text{e}}$$
LOADS AND LOAD DISTRIBUTIONS ON THE VERTICAL TAIL SURFACES DURING RUDDER KICKS AND FISHTAILS

In figure 22 (a), the rates of control deflection are shown plotted against airspeed for all rudder kicks made from the wings-level condition and in figure 23 (a) the rates are plotted against the maximum incremental pedal force. The faired lines in figure 23 (a) define the envelope of the maximum rate of control deflection attained. The maximum rate of deflection is noted to decrease with increase of pedal force, or amount of resistance to deflection. This result is in agreement with the results of tests made on the ground to determine the rates of elevator deflection used by a number of pilots (reference 7). On the basis of the relation indicated in figure 23 (a), the envelope describing the maximum rate (fig. 22 (a)) can be explained by the amount of resistance encountered. For instance, the rate of control deflection is greatest for the condition of power off and low speed.

In figure 24 (a) the ratio of rate of control deflection and amount of control deflection is plotted against speed for power on and power off. This figure shows that the ratio \( \frac{\Delta \delta}{\delta} \) approaches an upper limit of 10; the reciprocal of this ratio signifies that the minimum time to reach the highest control deflection the pilot can attain at each flight condition is a constant equal to 0.1 second. The conclusion that the ratio \( \frac{\Delta \delta}{\delta} \) is a constant may be deduced from the fact that both the maximum amount of deflection the pilot can attain \( \Delta \delta \), and the maximum rate of deflection \( \dot{\delta} \), are proportional to the same factor (the pedal force). It should be pointed out here that the rate of control deflection \( \dot{\delta} \), used in the ratio is the maximum measured during each rudder kick (see symbols) so that the minimum time value is derived from values of the ratio, which are themselves minimums.

Similar data obtained from the rudder kicks against an initial sideslip are presented superposed on the data obtained from kicks made from the wings-level condition in figures 22 (b), 23 (b), and 24 (b). It is shown in both figures 22 (b) and 23 (b) that the rates of deflection are higher than the maximums defined by the envelope for the data for rudder kicks from the wings-level condition. This result is obtained because the increment in pedal force is measured from the initial sideslip value, which in this case is an untrimmed value, so that a resistance to deflection is indicated that is higher than actually exists. Actually, the rudder tends to move toward the trim position of its own accord when the pilot releases it to apply opposite rudder. Figure 24 (b) shows that the time to reach the maximum rudder deflection is the same constant value as that obtained by rudder kicks from the wings-level condition. In this case, the greater rates are evidently balanced by a greater increment of control deflection.

Deflection load associated with maximum rate of control deflection.—The maximum deflection load per unit rudder deflection is shown plotted against dynamic pressure in figure 25 and is compared with the value computed from the geometric parameters of the tail for an infinite rate of deflection. The loads with power on are shown to be greater than the computed values at the lower speeds due to the fact that for the computed values the dynamic pressure at the tail was assumed to be equal to the free-stream dynamic pressure. At high speeds the actual maximum load experienced is almost 100 percent of that for an infinite rate of control deflection for this airplane.

For actual cases where the rate of deflection is between zero and infinity, the deflection load is dependent upon the rate of deflection, amount of deflection, and the response characteristics of the airplane.

For an airplane of given characteristics the amount of control deflection that can be applied and the response characteristics of the airplane are, in general, fixed so that it becomes convenient to consider the rate of control deflection as the prime determinant of the deflection load. The deflection load thus involves a determination of (1) the maximum rate of control deflection the pilot employs and (2) the load corresponding to this maximum rate.

Rate of control deflection.—From the many rudder kicks performed in this investigation some information was obtained which pertained to the rate at which the controls were deflected. It is to be emphasized that these are the rates that the pilot actually used, which may or may not be those of which he is physically capable.

Data pertaining to the maximum rate at which the pilot deflects the rudder is shown in figures 22 (a), 23 (a), and 24 (a) for kicks made from the wings-level condition and in figures 22 (b), 23 (b), and 24 (b) for kicks against an initial sideslip.

![Figure 14](image-url)
FIGURE 15.—Time histories of three abrupt rudder kicks to the left made at $V_r = 100$, 200, and 300 miles per hour with power on.

(a) Flight 6, run 2; $V_r = 100$ miles per hour.
(b) Flight 8b, run 2; $V_r = 200$ miles per hour.
(c) Flight 11a, run 3; $V_r = 300$ miles per hour.
As previously mentioned, the maximum rate of control deflection \( \Delta \delta_r = 0.1 \) is based upon the assumption of a linear-type control deflection which has a constant rate equal to the measured maximum rate. This assumed control deflection compared with a typical flight control deflection is shown in figure 26 (a). In figure 26 (b) the theoretical effect of rate of rudder movement on the deflection load is shown. The computations were made for the linear-type control deflection by the method indicated in reference 5. The figure shows the deflection load in percent of the load for an infinite rate of deflection \( \Delta \delta_r = 0 \) plotted against the time to reach maximum deflection \( \Delta \delta_r/\delta_r \). For the maximum rate of control deflection used by the pilot (a minimum time to reach maximum deflection of 0.1 sec) the load at 100 miles per hour is almost equal to that for an infinite rate of deflection. At higher speeds the rate becomes more critical in that the airplane responds more rapidly; however, even at a speed of 300 miles per hour the deflection load for a control deflection completed in 0.1 second is approximately 95 percent of that for an infinite rate.

**Dynamic Load**

**General relations.**—In figure 19 time histories of the component of load on the tail associated with the angular acceleration and the component due to sideslip are shown for one run, together with a comparison of the time histories of the summation of the components and the measured vertical-tail load. In figure 27 the measured dynamic loads are shown compared with the load computed from the relation

\[
\Delta N_x = \frac{dC_x}{d\beta} \Delta \rho q S \frac{b}{x_n} \frac{I_x}{x_n} \frac{\psi_2}{x_n}
\]

The data for rudder kicks against sideslip (fig. 27 (b)) are noted to have a slightly different slope from those of rudder kicks from the wings-level condition (fig. 27 (a)). The difference is presumed to be a result of differences in the action of secondary effects such as damping in roll or linear
acceleration. The comparisons, however, indicate that for the test airplane the equation adequately represents the dynamic-loads data. Thus the dynamic load following a rudder kick may be easily determined if the maximum value of sideslip $\Delta \beta$ and yawing acceleration $\dot{\psi}$ are available.

Some further discussion is needed regarding the factors which affect the angle of sideslip and the angular acceleration attained.

**Angle of sideslip.**—For steady sideslips the amount of sideslip attained by a given rudder angle is proportional to the factor $d\beta/d\delta$, (fig. 8). In abrupt rudder kicks, however, for an airplane with less than critical damping, a transitory angle of sideslip which is greater than the final steady sideslip will occur. For the case of zero directional damping and an abrupt rudder deflection, this transitory angle of sideslip would amount to twice the steady-state value of sideslip for the same rudder angle or $2(d\beta/d\delta)$.

The test airplane has low directional damping (as do most...
conventional airplanes) so that an overshoot resulting in a magnification factor of 1.5 to 2.0 over the steady-state value is to be expected. An approximate value of this factor for the test airplane may be obtained from figure 28 (a) which shows a plot of the ratio of angle of sideslip reached in rudder kicks to the value which would be reached in steady sideslips with the same rudder angle. At speeds of 100 and 200 miles per hour the full magnification factor is not reached because the rudder generally is reversed before the maneuver has continued long enough for the potential sideslip angle to be realized. The early rudder reversal relative to the time of maximum sideslip is shown in the time histories of the rudder kicks made at low speed (see fig. 13) and the computed effect of various times of rudder reversal on the sideslip reached is shown in figure 28 (b). At 300 miles per hour the rudder, in general, was held long enough for the full sideslip to be realized so that the magnification factor of approximately 1.5 obtained at this speed is believed to be near the true value for the test airplane.

Angular acceleration.—The maximum angular acceleration \( \dot{\psi}_2 \) is made up of the superposition of a component that is proportional to the amount of overshoot and a component resulting from the reversal of the rudder. The component due to the amount of overshoot depends upon the amount of damping, being zero for the case of critical damping and equal to the deflection angular acceleration \( \dot{\psi}_1 \) for zero damping. The component of angular acceleration due to rudder reversal is dependent upon the rate and amount of control deflection in the same manner as is the deflection angular acceleration. If the reversal deflection has the same rate and amount as the initial deflection (U-type rudder manipulation), the reversal component will exactly equal the deflection angular acceleration \( \dot{\psi}_2 \).

The two parts making up the yawing acceleration \( \dot{\psi}_2 \) are indicated in figure 29 in which the time histories of the load associated with the yawing acceleration only are shown for two rudder kicks in which the rudder was returned to zero after different time intervals. The time history for run 5 indicates the maximum angular acceleration without the reversal; whereas in run 6 the rudder was reversed at the time of maximum sideslip so that the maximum yawing acceleration includes the effect of rudder reversal. From this figure it is evident that the rudder kick in which the maneuver was stopped earlier results in higher loads because of the superposition of the two yawing-acceleration components near the time of their maximum values.
In order to indicate the likelihood with which the angular accelerations superpose at their maximum values, the ratio of the second peak angular acceleration to the first peak $\dot{\psi}_2/\dot{\psi}_1$ is shown plotted against speed in figure 30(a). In general, an approach of the ratio to a factor of 2 would indicate that the angular acceleration components superposed at their peaks; without the reversal component the ratio would be less than 1.0 since the overshoot component of $\dot{\psi}_2$ alone will always be less than the deflection value. Strictly speaking this value is obtained only for U-type control manipulation and, as indicated by some high values of the ratio (as high as 2.45), the rudder was returned past the trim position in some cases. The time histories (figs. 13 and 15) indicate, however, that although the rudder reversal was made at rates and amounts sometimes greater and sometimes less than the initial rudder kick the U-type manipulation represents an average type.

The computed effect of the time interval during which the rudder is held upon the manner of superposition of the angular acceleration components is shown in figure 30(b).

The data of figure 30(a) show that at 300 miles per hour the average of the components of angular acceleration due to overshoot and rudder reversal superpose near their maximum values and also that the U-type rudder manipulation is not an unduly conservative one as is sometimes felt in the specification of control motions.

**Estimate of maximum value for dynamic load from flight data.** An approximate formula for the estimation of the order of magnitude of the dynamic load would assist in assessing the relative significance of the factors involved. For this purpose the expression for the load on the vertical tail in terms of an effective angle of attack is most convenient; that is,

$$\Delta N_{v_2} = \Delta \alpha \left( \frac{dC_2}{d\alpha} \right)_v qS_v$$

$$= \left[ -\beta + \left( \frac{d\alpha}{d\delta_r} \right)_v \Delta \delta_r \right] \left( \frac{dC_2}{d\alpha} \right)_v qS_v$$

This expression is adequate when maximum values are considered inasmuch as the angular velocity is zero at the time of maximum $\beta$; also, the sidewise factor may be assumed to be zero.
LOADS AND LOAD DISTRIBUTIONS ON THE VERTICAL TAIL SURFACES DURING RUDDER KICKS AND FISHTAILS

(a) Rudder kicks from wings level.
(b) Rudder kicks against initial sideslip.

FIGURE 22—Rates of rudder deflection used by pilot plotted against equivalent airspeed with power on and power off.

(a) Rudder kicks from wings level.
(b) Rudder kicks against initial sideslip.

FIGURE 23—Rate of rudder deflection plotted against increment of pedal force used by pilot.
(a) Rudder kicks from wings level.

(b) Rudder kicks against initial sideslip

Figure 24.—Reciprocal of time to reach maximum rudder deflection against equivalent airspeed with power on and power off.

Figure 25.—Deflection load per degree rudder deflection plotted against dynamic pressure compared with deflection load for infinite rate of control deflection. Tailed symbols denote rudder kick against initial sideslip.
LOADS AND LOAD DISTRIBUTIONS ON THE VERTICAL TAIL SURFACES DURING RUDDER KICKS AND FISHTAILS

**Figure 26.** Computed deflection load on vertical tail of test airplane in percent of load for infinite rate of deflection against time to reach final control deflection and comparison of typical control deflection with linear type assumed for computations.

**Figure 27.** Comparison of measured dynamic load with dynamic load computed from the relation \( \Delta N_y = -\frac{dC_x}{d\delta} \Delta \delta \).
The angle of sideslip attained in a rudder kick may be written as

$$\beta = k \left( \frac{d\delta}{d\delta_r} \right) \Delta \delta,$$

where \( \frac{d\delta}{d\delta_r} \) is the measured slope as obtained from steady sideslips and \( k \) is a magnification factor which, as noted previously, would range from a value of 1 for a critically damped airplane to a value of 2 for zero damping. Thus,

$$\Delta N_{z} = k \left( \frac{dC_L}{d\alpha} \right) \Delta \delta q S + \left( \frac{dC_L}{d\delta} \right) \Delta \delta q S,$$

For the critical case of a rudder reversal at the time of maximum dynamic load the term \(-\left( \frac{dC_L}{d\delta} \right) \Delta \delta q S\) is added to the expression. If the reversal is assumed to be made at an infinite rate and to be equal to the initial deflection, the load becomes

$$\Delta N_{z} = -k \left( \frac{d\delta}{d\delta_r} \right) \left( \frac{dC_L}{d\alpha} \right) \Delta \delta q S.$$
LOADS AND LOAD DISTRIBUTIONS ON THE VERTICAL TAIL SURFACES DURING RUDDER KICKS AND FISHTAILS

Dynamic pressure, \( q \), lb/ft²

**FIGURE 31.** Dynamic balance load per degree rudder deflection plotted against dynamic pressure. Tailed symbols denote rudder kick against initial sideslip.

For the test airplane \( \frac{d\beta}{d\delta} \) is approximately equal to 1.5 (fig. 8) and as an upper-limit value, \( k = 2.0 \). The comparison of the measured load with the load computed from the approximate formula is shown by the line in figure 31.

**LOAD DISTRIBUTION**

In order to furnish a general picture of the distribution of load during a rudder kick, isometric views of the pressure distribution over the vertical tail during right and left rudder kicks are shown in figure 32. The figure shows the distributions on the vertical tail. The time of maximum deflection load, an intermediate point in the maneuver, and the time of maximum dynamic load. It can be seen from this figure and the time histories (figs. 14, 16, and 18) that the rudder carries most of the deflection load and that the fin carries most of the dynamic load. As regards the chordwise distribution of load, all types of distributions appear to occur during the rudder kick. The deflection load represents the zero yaw full rudder load; the intermediate point during the maneuver is the balance-type load; and the maximum dynamic load is a high angle of attack type of load with high leading-edge pressures.

**Distribution of load between rudder and fin.**—Further information on the distribution of the load between the rudder and fin is given in figures 33 and 34. A comparison of the magnitude of the deflection load on the rudder with that on the total vertical tail is shown in figure 33 (a) for rudder kicks from the wings-level condition and in figure 33 (b) for rudder kicks against initial sideslip. As shown by the time histories of figures 14 and 16 the maximum deflection load on the rudder occurs after the maximum on the total vertical tail so that the load values plotted in figure 33 do not necessarily occur at the same time. From figure 33, the load on the rudder is found to be approximately equal to the total deflection load. For the high loads which were attained at 300 miles per hour the rudder deflection load is actually greater than that on the total vertical tail. This condition results from a combination of the lower rate of control deflection with the more rapid airplane response, with the consequence that the airplane starts to yaw before the rudder has completed its travel. The yawing velocity imposes a load on the fin that is opposite to the rudder load and results in a lower net load on the tail. This effect is illustrated in figure 32 by the higher pressures on the rudder at an intermediate point during the maneuver rather than at the time of maximum vertical-tail deflection load.

A comparison of the dynamic load carried by the fin with that carried by the total vertical tail is shown in figure 34 (a) for rudder kicks from the wings-level condition and in figure 34 (b) for rudder kicks against steady sideslip. The fin is shown to carry approximately 90 percent of the dynamic load in rudder kicks from the wings-level condition and about 100 percent of the dynamic load in kicks against sideslip. When the fin carries a load greater than 100 percent, the total load includes a rudder load in a direction opposite to that on the fin.
Spanwise and chordwise load distribution.—The spanwise load distributions on the fin at the time of maximum fin load and on the rudder at the time of maximum rudder load are presented in figure 35 for power on and figure 37 for power off for the most severe rudder kicks made in each direction and at each test speed. The symbols in these figures are used to distinguish chordwise-load points of two runs having approximately the same value of load. The chordwise pressure distributions over rib V (fig. 5) obtained at times corresponding to the times for which the spanwise load distributions are shown are presented in figures 36 and 38.

Figure 39 shows that the spanwise center of load on the fin varies slightly depending upon the direction of kick as well as upon the airspeed. On an average, the spanwise center of load is 10 percent farther outboard than the airload distribution for which the surfaces were designed.

The chordwise load distributions in figures 36 and 38 show that (except at an airspeed of 100 mph) the maximum fin load is, in general, associated with a small value of load on the rudder, whereas the maximum rudder load occurs during an intermediate point in the maneuver when the fin has some load due to yawing.

LOAD DIAGRAMS

The construction of load diagrams for the vertical tail surfaces may be made by the use of the foregoing results. For instance, the deflection load was shown to be critical for the rudder. At high speeds the total deflection load was less than the load for an infinite rate of control deflection (see fig. 25) but the load on the rudder was greater than 100 percent of the deflection load, and it is therefore reasonable to assume that the critical rudder load may be equal to the total deflection load at an infinite rate of control deflection. Thus,

$$\Delta N_r = \Delta \delta_r \left( \frac{dC_l}{d\delta_r} \right) q S_r$$

In figure 40 (a) the load computed by this equation is shown to compare well with the maximum values of measured rudder loads.

The dynamic load was found to be critical for the fin. The load on the fin may be expressed as some fraction $K$ of
the dynamic load. The factor $K$ may be determined from the geometric characteristics of the tail for the assumption of a hypothetical control motion in which the rudder is returned to zero at the time of maximum sideslip; that is,

$$\Delta N_2 = K \Delta N_3$$

$$= K \left[ -k \frac{d\beta}{d\delta_e} \left( \frac{dC_l}{d\alpha} \right) \Delta S \right]$$

For the test airplane the factor $K$ for this condition was shown to be 90 percent in rudder kicks from the wings-level condition (fig. 34 (a)).

In figure 40 this relation is shown on the basis of the load per degree rudder deflection against dynamic pressure along with experimental values. In the calculations the magnification factor $k$ was assumed to be 2.0 and $\frac{d\beta}{d\delta_e} = 1.5$.

The load diagram in figure 41 was constructed from the preceding formulas. The dashed lines show computed loads.
for two pedal forces and the points represent the largest experimental values obtained at equivalent airspeeds of 200 and 300 miles per hour.

RESULTS AND DISCUSSION—FISHTAIL MANEUVERS

Vertical-tail failures have occurred on military airplanes during evasive action or fishtail maneuvers. Some concern has therefore been expressed about including the fishtail maneuver as a critical design condition because the weight penalty for adequate strength was considered prohibitive. In addition, there was for a time an impression among some designers that the vertical tail could fail on any airplane if the rudder were deflected in a sinusoidal manner at the natural frequency of the airplane. Consequently, it seemed to be in order that a specification be made as to how far the maneuver was to be continued. For this purpose, an analogous system which is familiar in simple dynamics may be used to furnish useful information concerning the fishtail maneuver.

FIGURE 36.—Chordwise pressure distributions over rib V (see fig. 5) for spanwise load distributions of figure 35.
LOADS AND LOAD DISTRIBUTIONS ON THE VERTICAL TAIL SURFACES DURING RUDDER KICKS AND FISHTAILS

DISCUSSION OF RESULTS

As was pointed out in reference 5, the fishtail maneuver can be assumed to be a flat yawing maneuver so that the solution to this problem might be equivalent to that for a linear single-spring system. A brief review of well-known results of the spring system from simple dynamics will therefore furnish a useful background. The curves shown in figure 42 (taken from reference 8) apply to the case of an external sinusoidal force acting upon the spring system.

Figure 42 (a) shows the amplitude magnification factor plotted against the ratio of the frequency of the impressed force to the natural frequency of the system for systems having different ratios of damping to critical damping. In figure 42 (b) the phase relation between the impressed force and the amplitude is presented for the same conditions. In terms of what happens in the fishtail maneuvers the following observations may be made from this figure.

(1) For an airplane with some damping the sideslip (or amplitude) magnification will reach a finite equilibrium value even for the case of a rudder oscillation having the same frequency as the airplane. The amount of magnification is dependent upon the ratio of the damping to the critical damping and, of course, upon the frequency at which the rudder is deflected relative to the natural frequency of the airplane.

(2) The rudder angle (or forcing function) is out of phase with the angle of sideslip (or amplitude) by an amount depending upon the amount of relative damping. At resonance, however, the phase relation is always 90°. For resonance, therefore, for a perfect fishtail, the rudder angle will be zero at the time of maximum sideslip and maximum at the point of zero sideslip.

It should be noted at this point that these curves could have been derived in terms of loads in which case the magnifications of figure 42 (a) would then be expressed in terms of load magnification. For the case where the impressed frequency is the same as the airplane frequency, in which case the rudder deflection would be zero at the time of maximum sideslip (fig. 42 (b)), the expression for the load in a fishtail maneuver would become

\[ \Delta N_y = -\beta \left( \frac{dC_{\alpha}}{d\alpha} \right)_x g S_e \]

ANALYSIS OF TESTS

The results obtained during the fishtail investigation are given in table V. The first eight of these fishtails were slightly artificial since the pilot deliberately tried to obtain high tail loads, whereas the last four were made in as natural and comfortable a manner as possible.

The first set of maneuvers was intended to show how critical the maneuver could be if the pilot deliberately tried to work the rudder control at the same frequency as the airplane frequency in order to reach high angles of yaw. The time histories of these maneuvers are presented in figures 43 and 44 for the power-on and power-off maneuvers made at 150 and 200 miles per hour, respectively. In figure 45 are presented power-on and power-off fishtail maneuvers in which the pilot kicked the rudder against the swing at the point of maximum yawing velocity. All of these maneuvers (figs. 43 to 45) were very uncomfortable to the pilot because of the severe pitching which resulted.
The second set of tests consisted of the fishtail maneuvers in which a different pilot performed a mild fishtail maneuver in as comfortable a manner as possible. These maneuvers are presented in figures 46 and 47 at speeds of 200 and 250 miles per hour and 300 and 350 miles per hour, respectively.

A study of the time histories of the fishtail maneuvers yields the following deductions:

(1) The maneuvers in which the pilot was free to coordinate the controls show that the pitching was very much less, with the result that the maneuver was not particularly uncomfortable.

(2) Within only one cycle of rudder motion the loads attain values close to the maximum measured during the whole maneuver.

(3) As the maneuver continues, the load on the rudder tends to bear the $90^\circ$ phase relation with the load on the fin. This result is indicated in figure 42 (b) for the condition of resonance.
(4) The abrupt rudder deflection applied against the maximum velocity of swing results in high rudder loads (fig. 43). If the rudder is moved against the airplane swing, the phase relation of the rudder and fin loads is disturbed so that the loads become additive.

Frequency of rudder operation with relation to frequency of airplane.—One of the points of interest in the fishtail tests was to note whether, as might be expected, the pilot tends to move the rudder in phase with the airplane frequency. In order to obtain the average rudder frequency for each maneuver, the actual control manipulation was arbitrarily approximated by a sine function. The rudder control deflections for all 12 runs are shown in figure 48 in nondimensional form; the actual control deflection was divided by the amplitude of the sine curve used in the approximation of the motion. The assumed sine curves are also shown. The natural frequency \( f_n \) of the airplane was computed from the expression

\[
f_n = \frac{1}{2\pi} \sqrt{\frac{K_2}{K_1}}
\]

where \( K_1 \) and \( K_2 \) are determined from the aerodynamic characteristics of the airplane and are defined by equation (5) of reference 5.

Inasmuch as the period \( 1/f_n \) is a more usual way of plotting the airplane response, the data are shown plotted in that manner in figure 49. From this figure it is seen that the fishtail maneuvers made by the pilot when his actions were unrestricted (symbols with tails) were as close to the airplane period as those maneuvers in which he attempted to work the controls at the same period as the airplane. Although the control deflections are irregular, the results indicate that the pilot does tend to work the controls in phase with the airplane frequency in performing a fishtail.
(a) Flight 16, run 1; power on.
(b) Flight 16, run 2; power off.

**Figure 43.**—Time histories of measurements recorded during power-on and power-off fishtails at 150 miles per hour.
FLIGHT 25A, RUN 1; POWER ON.
FLIGHT 25A, RUN 2; POWER OFF.

FIGURE 44. — Time histories of measurements recorded during power-on and power-off fishballs at 200 miles per hour.
Yew on gulor

(a) Flight Ma, run I; powrr on. (b) Flight 18n, run 3: power off.

FIGURE 45.—Time histories of measurements recorded during power-on and power-off fishtails at 180 miles per hour in which pilot kicked rudder against the swing at point of maximum yawing velocity.
Figure 46.—Time histories of measurements recorded during mild fishtail maneuvers at 200 and 250 miles per hour with power on.

(a) Flight 26, run 1; \( V_a = 200 \) miles per hour.
(b) Flight 26, run 2; \( V_a = 250 \) miles per hour.

(a) Flight 26, run 1; \( V_a = 200 \) miles per hour.
(b) Flight 26, run 2; \( V_a = 250 \) miles per hour.

LOADS AND LOAD DISTRIBUTIONS ON THE VERTICAL TAIL SURFACES DURING RUDDER KICKS AND FISHTAILS
FIGURE 47.—Time histories of measurements recorded during mild fishtail maneuvers at 300 and 350 miles per hour with power on.

(a) Flight 26, run 3; $V_a=300$ miles per hour.

(b) Flight 26, run 4; $V_a=350$ miles per hour.
Figure 48.—Comparison of rudder manipulations of all fishtails with sine curves.
Comparisons between measured and estimated load.—A comparison of the measured loads with those computed on the basis of the theory of flat yawing (reference 5) is presented in figure 50, which shows the maximum tail load measured per degree of rudder deflection during each run. Mean amplitudes of rudder deflection were used to obtain the experimental values of load per degree. Also included in figure 50 is a line corresponding to the load per degree for a control motion in which the rudder was assumed to be returned to trim at the time of maximum sideslip. Figure 50 shows that the loads measured during the fishtail did not reach the computed resonant value but were more nearly equal to the values given by the equation representing the hypothetical U-type control motion.

LOAD DISTRIBUTIONS

The fishtail maneuvers, as indicated by simple dynamics, yield an angle-of-attack load with rudder at zero deflection plus a zero-yaw full-rudder load according to the phase relations indicated by figure 42 (b).

Figure 51 presents the spanwise load distributions over the rudder and fin at various times during the power-on fishtail maneuvers of figures 43, 44, and 45. The spanwise and chordwise load distributions over fin and rudder and chordwise load distributions over rib V during the fishtails of figures 46 and 47 are presented in figures 52 and 53, respectively. Figure 54 (a) presents the center of load on the fin at the times of maximum loads on the fin during the fishtail.
(a) Flight 16, run 1; high yaw angle; \( V_r = 150 \) miles per hour.

(b) Flight 18a, run 1; high yaw angle; \( V_r = 200 \) miles per hour.

(c) Flight 18a, run 1; kick against swing; \( V_r = 150 \) miles per hour.

Figure 51. — Spanwise load distributions over the rudder and fin at various times during the power-on fishtails of figures 43, 44, and 45.
Also, for illustrative purposes, time histories of the center-of-load variation during the fishtails of figures 46 and 47 are presented in figure 54 (b).

CONCLUSIONS

The conclusions are grouped under the general subject heading from which they were derived.

First load peak following a rudder kick (deflection load)

1. The deflection load can be determined with sufficient accuracy by the product of the moment of inertia and the first maximum yawing acceleration divided by the tail length.
2. The minimum time used by the pilot to attain the maximum rudder deflection at each flight condition appears to be a constant.
3. The deflection load on the vertical tail of the test airplane reaches values close to those for an infinite rate of control deflection.

Second load peak following a rudder kick (dynamic load)

1. The dynamic load can be determined with sufficient accuracy by the sum of the component of load necessary to balance the unstable yawing moment of the wing-fuselage combination in sideslip and the component of load due to angular acceleration in yaw.
2. After the initial rudder kick the return of the rudder to trim was, in general, made at the time of maximum sideslip so that the load due to abrupt reversal of the rudder was superimposed at the time of maximum overshoot load.
3. A rational approximate formula based upon a U-type control deflection satisfactorily expresses the upper limit value of the measured dynamic loads for this airplane. This formula is in terms of the sideslip-rudder ratio from steady-flight results and a magnification factor which considers the amount of directional damping in the airplane.
Load distributions

1. The critical loads on the rudder are associated with the deflection load. The deflection load on the rudder is approximately equal to the total deflection load on the tail.

2. The critical loads on the fin are associated with the dynamic load on the tail. The upper limit of the measured dynamic loads on the fin is satisfactorily expressed as the fraction of the total dynamic load which would be carried for the rudder at zero.

3. At the time of maximum fin load the spanwise center of load on the fin is 10 percent farther outboard than the design air-load distribution.
LOADS AND LOAD DISTRIBUTIONS ON THE VERTICAL TAIL SURFACES DURING RUDDER KICKS AND FISHTAILS

(a) Center of pressure at time of maximum load.
(b) Time history of center of pressure.

Fishtail maneuvers

1. The maximum loads measured during the fishtail maneuvers were no greater than those which would result from a hypothetical U-type rudder kick in which the rudder is returned to zero at the time of maximum sideslip.

2. As might be expected, the pilot tends to work the rudder in phase with the natural frequency of the airplane.

3. At resonance, the rudder angle and sideslip angle are 90° out of phase so that at maximum sideslip the rudder deflection is zero and the load is proportional to the sideslip angle.

4. An abrupt stopping action in which the rudder is kicked against the swing results in high rudder loads. If the control is worked against the airplane swing, the phase relation between the rudder and fin loads is disturbed so that the loads become additive.

LANGLEY MEMORIAL AERONAUTICAL LABORATORY,
NATIONAL ADVISORY COMMITTEE FOR AERONAUTICS,
LANGLEY FIELD, VA., April 9, 1947.

REFERENCES


TABLE I.—GEOMETRIC CHARACTERISTICS

<table>
<thead>
<tr>
<th>Wing:</th>
<th>236</th>
</tr>
</thead>
<tbody>
<tr>
<td>Area, sq ft</td>
<td></td>
</tr>
<tr>
<td>Span, ft</td>
<td>37.29</td>
</tr>
<tr>
<td>Wing area</td>
<td></td>
</tr>
<tr>
<td>Mean aerodynamic chord, ft</td>
<td>0.8</td>
</tr>
<tr>
<td>Root chord, ft</td>
<td>9</td>
</tr>
<tr>
<td>Section at root</td>
<td>NACA 2215</td>
</tr>
<tr>
<td>Section at tip</td>
<td>NACA 2209</td>
</tr>
<tr>
<td>Angle to thrust line, deg</td>
<td>1</td>
</tr>
<tr>
<td>Dihedral, deg</td>
<td>6</td>
</tr>
<tr>
<td>Aspect ratio</td>
<td>5.9</td>
</tr>
<tr>
<td>Engine:</td>
<td>Allison V-1710-F4R</td>
</tr>
<tr>
<td>Normal power at 10,800 ft, hp</td>
<td>1000</td>
</tr>
<tr>
<td>Propeller gear ratio</td>
<td>2:1</td>
</tr>
<tr>
<td>Propeller diameter, ft</td>
<td>11</td>
</tr>
<tr>
<td>Flight operation:</td>
<td></td>
</tr>
<tr>
<td>Average weight in flight, lb</td>
<td>8200</td>
</tr>
<tr>
<td>Average position, percent M. A. C.</td>
<td>29.5</td>
</tr>
<tr>
<td>Vertical tail surface:</td>
<td></td>
</tr>
<tr>
<td>Total area, sq ft</td>
<td>22.9</td>
</tr>
<tr>
<td>Height above fuselage, ft</td>
<td>5.67</td>
</tr>
<tr>
<td>Fin area (less fairing area), sq ft</td>
<td>9.18</td>
</tr>
<tr>
<td>Rudder area (including 1.94 sq ft of balance and 0.55 sq ft of tab), sq ft</td>
<td>13.74</td>
</tr>
<tr>
<td>Distance from c. g. to rudder hinge line, ft</td>
<td>20.13</td>
</tr>
<tr>
<td>Fin offset, deg</td>
<td>0</td>
</tr>
<tr>
<td>Horizontal tail surface:</td>
<td></td>
</tr>
<tr>
<td>Total area, sq ft</td>
<td>48.3</td>
</tr>
<tr>
<td>Span, ft</td>
<td>12.79</td>
</tr>
<tr>
<td>Stabilizer area (including 3.54 sq ft of fuselage), sq ft</td>
<td>30.86</td>
</tr>
<tr>
<td>Elevator area (including 3.8 sq ft of balance and 1.68 sq ft of tab), sq ft</td>
<td>17.44</td>
</tr>
<tr>
<td>Distance from wing root L. E. to elevator hinge line, ft</td>
<td>20.0</td>
</tr>
<tr>
<td>Stabilizer set above thrust line, deg</td>
<td>2</td>
</tr>
<tr>
<td>Horizontal tail above fuselage center line, ft</td>
<td>1.50</td>
</tr>
<tr>
<td>Maximum elevator deflection (up), deg</td>
<td>31.5</td>
</tr>
</tbody>
</table>
**TABLE II—STEADY-SIDESLIP TESTS**

<table>
<thead>
<tr>
<th>Run</th>
<th>$V_0$ (mph)</th>
<th>$\Delta \theta$ (°)</th>
<th>$T_\alpha$</th>
<th>$Q_\alpha$</th>
<th>$\Delta \phi$ (deg)</th>
<th>$\Delta \psi$ (deg)</th>
<th>$\Delta N_y$ (lb)</th>
<th>$\Delta N_x$ (lb)</th>
<th>$\Delta C_{N_y}$</th>
<th>$\Delta C_{N_x}$</th>
<th>$\Delta C_{N_z}$</th>
</tr>
</thead>
<tbody>
<tr>
<td>3</td>
<td>105.0</td>
<td>-11.49</td>
<td>0.109</td>
<td>0.007</td>
<td>-5.25</td>
<td>-2.90</td>
<td>162</td>
<td>-1</td>
<td>151</td>
<td>0.196</td>
<td>-0.002</td>
</tr>
<tr>
<td>4</td>
<td>150.0</td>
<td>-9.96</td>
<td>0.109</td>
<td>0.007</td>
<td>7.65</td>
<td>1.90</td>
<td>-162</td>
<td>25</td>
<td>-149</td>
<td>-248</td>
<td>-0.040</td>
</tr>
<tr>
<td>5</td>
<td>160.0</td>
<td>-12.00</td>
<td>0.109</td>
<td>0.007</td>
<td>-10.30</td>
<td>-2.69</td>
<td>260</td>
<td>-23</td>
<td>223</td>
<td>129</td>
<td>-0.106</td>
</tr>
<tr>
<td>6</td>
<td>165.0</td>
<td>-10.15</td>
<td>0.044</td>
<td>-0.004</td>
<td>-6.85</td>
<td>-1.37</td>
<td>358</td>
<td>124</td>
<td>-241</td>
<td>361</td>
<td>-0.014</td>
</tr>
<tr>
<td>11</td>
<td>165.0</td>
<td>-3.20</td>
<td>0.044</td>
<td>-0.004</td>
<td>-4.24</td>
<td>-1.23</td>
<td>323</td>
<td>43</td>
<td>-155</td>
<td>-71.5</td>
<td>0.007</td>
</tr>
<tr>
<td>12</td>
<td>210.3</td>
<td>-3.20</td>
<td>0.044</td>
<td>-0.004</td>
<td>-2.29</td>
<td>-1.32</td>
<td>273</td>
<td>-54</td>
<td>221</td>
<td>0.658</td>
<td>-0.010</td>
</tr>
<tr>
<td>13</td>
<td>210.3</td>
<td>-7.59</td>
<td>0.044</td>
<td>-0.004</td>
<td>-1.41</td>
<td>-0.97</td>
<td>213</td>
<td>42</td>
<td>-115</td>
<td>-0.007</td>
<td>0.015</td>
</tr>
<tr>
<td>18</td>
<td>210.3</td>
<td>-3.15</td>
<td>0.044</td>
<td>-0.004</td>
<td>-3.00</td>
<td>-2.12</td>
<td>167</td>
<td>42</td>
<td>-115</td>
<td>-0.007</td>
<td>0.015</td>
</tr>
<tr>
<td>19</td>
<td>210.3</td>
<td>-6.55</td>
<td>0.044</td>
<td>-0.004</td>
<td>-4.61</td>
<td>-3.22</td>
<td>267</td>
<td>-62</td>
<td>-155</td>
<td>-0.006</td>
<td>0.014</td>
</tr>
<tr>
<td>20</td>
<td>210.3</td>
<td>-3.20</td>
<td>0.044</td>
<td>-0.004</td>
<td>-2.29</td>
<td>-1.32</td>
<td>273</td>
<td>43</td>
<td>-155</td>
<td>-71.5</td>
<td>0.007</td>
</tr>
<tr>
<td>21</td>
<td>275.5</td>
<td>-3.50</td>
<td>0.023</td>
<td>-0.004</td>
<td>-1.05</td>
<td>-0.77</td>
<td>180</td>
<td>-26</td>
<td>148</td>
<td>-0.009</td>
<td>0.017</td>
</tr>
<tr>
<td>22</td>
<td>275.5</td>
<td>-4.35</td>
<td>0.023</td>
<td>-0.004</td>
<td>-5.55</td>
<td>-3.33</td>
<td>231</td>
<td>-114</td>
<td>229</td>
<td>-0.025</td>
<td>0.009</td>
</tr>
<tr>
<td>23</td>
<td>275.5</td>
<td>-1.15</td>
<td>0.023</td>
<td>-0.004</td>
<td>-1.65</td>
<td>-0.97</td>
<td>231</td>
<td>-114</td>
<td>229</td>
<td>-0.025</td>
<td>0.009</td>
</tr>
<tr>
<td>24</td>
<td>275.5</td>
<td>-7.95</td>
<td>0.023</td>
<td>-0.004</td>
<td>-1.65</td>
<td>-0.97</td>
<td>231</td>
<td>-114</td>
<td>229</td>
<td>-0.025</td>
<td>0.009</td>
</tr>
<tr>
<td>25</td>
<td>275.5</td>
<td>-1.65</td>
<td>0.023</td>
<td>-0.004</td>
<td>-1.65</td>
<td>-0.97</td>
<td>231</td>
<td>-114</td>
<td>229</td>
<td>-0.025</td>
<td>0.009</td>
</tr>
<tr>
<td>26</td>
<td>275.5</td>
<td>-1.65</td>
<td>0.023</td>
<td>-0.004</td>
<td>-1.65</td>
<td>-0.97</td>
<td>231</td>
<td>-114</td>
<td>229</td>
<td>-0.025</td>
<td>0.009</td>
</tr>
<tr>
<td>27</td>
<td>275.5</td>
<td>-1.65</td>
<td>0.023</td>
<td>-0.004</td>
<td>-1.65</td>
<td>-0.97</td>
<td>231</td>
<td>-114</td>
<td>229</td>
<td>-0.025</td>
<td>0.009</td>
</tr>
<tr>
<td>28</td>
<td>275.5</td>
<td>-1.65</td>
<td>0.023</td>
<td>-0.004</td>
<td>-1.65</td>
<td>-0.97</td>
<td>231</td>
<td>-114</td>
<td>229</td>
<td>-0.025</td>
<td>0.009</td>
</tr>
<tr>
<td>29</td>
<td>275.5</td>
<td>-1.65</td>
<td>0.023</td>
<td>-0.004</td>
<td>-1.65</td>
<td>-0.97</td>
<td>231</td>
<td>-114</td>
<td>229</td>
<td>-0.025</td>
<td>0.009</td>
</tr>
<tr>
<td>30</td>
<td>275.5</td>
<td>-1.65</td>
<td>0.023</td>
<td>-0.004</td>
<td>-1.65</td>
<td>-0.97</td>
<td>231</td>
<td>-114</td>
<td>229</td>
<td>-0.025</td>
<td>0.009</td>
</tr>
<tr>
<td>31</td>
<td>275.5</td>
<td>-1.65</td>
<td>0.023</td>
<td>-0.004</td>
<td>-1.65</td>
<td>-0.97</td>
<td>231</td>
<td>-114</td>
<td>229</td>
<td>-0.025</td>
<td>0.009</td>
</tr>
<tr>
<td>32</td>
<td>275.5</td>
<td>-1.65</td>
<td>0.023</td>
<td>-0.004</td>
<td>-1.65</td>
<td>-0.97</td>
<td>231</td>
<td>-114</td>
<td>229</td>
<td>-0.025</td>
<td>0.009</td>
</tr>
<tr>
<td>33</td>
<td>275.5</td>
<td>-1.65</td>
<td>0.023</td>
<td>-0.004</td>
<td>-1.65</td>
<td>-0.97</td>
<td>231</td>
<td>-114</td>
<td>229</td>
<td>-0.025</td>
<td>0.009</td>
</tr>
<tr>
<td>34</td>
<td>275.5</td>
<td>-1.65</td>
<td>0.023</td>
<td>-0.004</td>
<td>-1.65</td>
<td>-0.97</td>
<td>231</td>
<td>-114</td>
<td>229</td>
<td>-0.025</td>
<td>0.009</td>
</tr>
<tr>
<td>35</td>
<td>275.5</td>
<td>-1.65</td>
<td>0.023</td>
<td>-0.004</td>
<td>-1.65</td>
<td>-0.97</td>
<td>231</td>
<td>-114</td>
<td>229</td>
<td>-0.025</td>
<td>0.009</td>
</tr>
<tr>
<td>36</td>
<td>275.5</td>
<td>-1.65</td>
<td>0.023</td>
<td>-0.004</td>
<td>-1.65</td>
<td>-0.97</td>
<td>231</td>
<td>-114</td>
<td>229</td>
<td>-0.025</td>
<td>0.009</td>
</tr>
</tbody>
</table>

*Initial steady-flight value (increment from wings-level trim).
### TABLE III.—RUDDER KICKS

| Flight | Run | V₁ (mph) | Power | θ₀ (deg) | T₀ | Q₀ | L₀ (lb) | θ₁ (deg) | φ₁ rad/sec | φ₀ rad/sec | α₀ rad/see | α₁ rad/see | α₀ (sec) |
|--------|-----|----------|-------|----------|----|----|---------|----------|------------|------------|-------------|-------------|-------------|-----------|
| 6      | 29  | 180      | On    | 0        | 0.76| 0.07| 6.030   | 0.232    | 0.200      | 0.200      | 0.200       | 0.200       | 0.200       | 0.200     |
| 7      | 30  | 180      | Rated | 0.017    | 0.76| 0.07| 6.030   | 0.232    | 0.200      | 0.200      | 0.200       | 0.200       | 0.200       | 0.200     |
| 8      | 31  | 180      | Rated | 0.017    | 0.76| 0.07| 6.030   | 0.232    | 0.200      | 0.200      | 0.200       | 0.200       | 0.200       | 0.200     |
| 9      | 32  | 180      | On    | 0        | 0.76| 0.07| 6.030   | 0.232    | 0.200      | 0.200      | 0.200       | 0.200       | 0.200       | 0.200     |
| 10     | 33  | 180      | Rated | 0.017    | 0.76| 0.07| 6.030   | 0.232    | 0.200      | 0.200      | 0.200       | 0.200       | 0.200       | 0.200     |
| 11     | 34  | 180      | Rated | 0.017    | 0.76| 0.07| 6.030   | 0.232    | 0.200      | 0.200      | 0.200       | 0.200       | 0.200       | 0.200     |
| 12     | 35  | 180      | On    | 0        | 0.76| 0.07| 6.030   | 0.232    | 0.200      | 0.200      | 0.200       | 0.200       | 0.200       | 0.200     |
| 13     | 36  | 180      | Rated | 0.017    | 0.76| 0.07| 6.030   | 0.232    | 0.200      | 0.200      | 0.200       | 0.200       | 0.200       | 0.200     |
| 14     | 37  | 180      | Rated | 0.017    | 0.76| 0.07| 6.030   | 0.232    | 0.200      | 0.200      | 0.200       | 0.200       | 0.200       | 0.200     |
| 15     | 38  | 180      | On    | 0        | 0.76| 0.07| 6.030   | 0.232    | 0.200      | 0.200      | 0.200       | 0.200       | 0.200       | 0.200     |
| 16     | 39  | 180      | Rated | 0.017    | 0.76| 0.07| 6.030   | 0.232    | 0.200      | 0.200      | 0.200       | 0.200       | 0.200       | 0.200     |

* Initial steady-flight value (increment from wings-level trim).
**TABLE IV.—RUDDER KICKS (IN WHICH NO LOADS WERE MEASURED)**

<table>
<thead>
<tr>
<th>Flight</th>
<th>Run</th>
<th>$V_a$ (mph)</th>
<th>Power</th>
<th>$\alpha_p$ (deg)</th>
<th>$\beta$ (deg)</th>
<th>$T_a$ (lb)</th>
<th>$\dot{T}_a$ (deg/sec)</th>
<th>$\dot{\beta}$ (deg/sec)</th>
<th>$\dot{\alpha}_p$ (deg/sec)</th>
<th>$\beta_p$ (deg)</th>
<th>$\alpha_p$ (deg)</th>
<th>$\omega$ (deg/sec)</th>
</tr>
</thead>
<tbody>
<tr>
<td>1</td>
<td>100</td>
<td>100</td>
<td>Rated</td>
<td>0.007</td>
<td>0.000</td>
<td>3.56</td>
<td>9.79</td>
<td>9.33</td>
<td>1.00</td>
<td>18.1</td>
<td>-1.30</td>
<td>12.00</td>
</tr>
<tr>
<td>2</td>
<td>100</td>
<td>100</td>
<td>Rated</td>
<td>0.007</td>
<td>0.000</td>
<td>3.56</td>
<td>9.79</td>
<td>9.33</td>
<td>1.00</td>
<td>18.1</td>
<td>-1.30</td>
<td>12.00</td>
</tr>
<tr>
<td>3</td>
<td>100</td>
<td>100</td>
<td>Rated</td>
<td>0.007</td>
<td>0.000</td>
<td>3.56</td>
<td>9.79</td>
<td>9.33</td>
<td>1.00</td>
<td>18.1</td>
<td>-1.30</td>
<td>12.00</td>
</tr>
<tr>
<td>4</td>
<td>100</td>
<td>100</td>
<td>Rated</td>
<td>0.007</td>
<td>0.000</td>
<td>3.56</td>
<td>9.79</td>
<td>9.33</td>
<td>1.00</td>
<td>18.1</td>
<td>-1.30</td>
<td>12.00</td>
</tr>
<tr>
<td>5</td>
<td>100</td>
<td>100</td>
<td>Rated</td>
<td>0.007</td>
<td>0.000</td>
<td>3.56</td>
<td>9.79</td>
<td>9.33</td>
<td>1.00</td>
<td>18.1</td>
<td>-1.30</td>
<td>12.00</td>
</tr>
<tr>
<td>6</td>
<td>100</td>
<td>100</td>
<td>Rated</td>
<td>0.007</td>
<td>0.000</td>
<td>3.56</td>
<td>9.79</td>
<td>9.33</td>
<td>1.00</td>
<td>18.1</td>
<td>-1.30</td>
<td>12.00</td>
</tr>
<tr>
<td>7</td>
<td>100</td>
<td>100</td>
<td>Rated</td>
<td>0.007</td>
<td>0.000</td>
<td>3.56</td>
<td>9.79</td>
<td>9.33</td>
<td>1.00</td>
<td>18.1</td>
<td>-1.30</td>
<td>12.00</td>
</tr>
<tr>
<td>8</td>
<td>100</td>
<td>100</td>
<td>Rated</td>
<td>0.007</td>
<td>0.000</td>
<td>3.56</td>
<td>9.79</td>
<td>9.33</td>
<td>1.00</td>
<td>18.1</td>
<td>-1.30</td>
<td>12.00</td>
</tr>
<tr>
<td>9</td>
<td>100</td>
<td>100</td>
<td>Rated</td>
<td>0.007</td>
<td>0.000</td>
<td>3.56</td>
<td>9.79</td>
<td>9.33</td>
<td>1.00</td>
<td>18.1</td>
<td>-1.30</td>
<td>12.00</td>
</tr>
<tr>
<td>10</td>
<td>100</td>
<td>100</td>
<td>Rated</td>
<td>0.007</td>
<td>0.000</td>
<td>3.56</td>
<td>9.79</td>
<td>9.33</td>
<td>1.00</td>
<td>18.1</td>
<td>-1.30</td>
<td>12.00</td>
</tr>
<tr>
<td>11</td>
<td>100</td>
<td>100</td>
<td>Rated</td>
<td>0.007</td>
<td>0.000</td>
<td>3.56</td>
<td>9.79</td>
<td>9.33</td>
<td>1.00</td>
<td>18.1</td>
<td>-1.30</td>
<td>12.00</td>
</tr>
<tr>
<td>12</td>
<td>100</td>
<td>100</td>
<td>Rated</td>
<td>0.007</td>
<td>0.000</td>
<td>3.56</td>
<td>9.79</td>
<td>9.33</td>
<td>1.00</td>
<td>18.1</td>
<td>-1.30</td>
<td>12.00</td>
</tr>
</tbody>
</table>

* Initial steady-flight value (increment from wings-level trim).

**TABLE V.—FISHTAIL MANEUVERS**

<table>
<thead>
<tr>
<th>Flight</th>
<th>Run</th>
<th>Speed (mph)</th>
<th>Power</th>
<th>$L$ (deg)</th>
<th>$\alpha_p$ (deg)</th>
<th>$\alpha_p$ (deg)</th>
<th>$\omega$ (deg/sec)</th>
</tr>
</thead>
<tbody>
<tr>
<td>1</td>
<td>150</td>
<td>On...</td>
<td>Rated</td>
<td>8.0</td>
<td>9.0</td>
<td>-900</td>
<td>-690</td>
</tr>
<tr>
<td>2</td>
<td>151</td>
<td>Off</td>
<td>Rated</td>
<td>13.5</td>
<td>14.0</td>
<td>1570</td>
<td>1250</td>
</tr>
<tr>
<td>3</td>
<td>152</td>
<td>On...</td>
<td>Rated</td>
<td>13.0</td>
<td>11.5</td>
<td>-790</td>
<td>-700</td>
</tr>
<tr>
<td>4</td>
<td>153</td>
<td>Off</td>
<td>Rated</td>
<td>13.0</td>
<td>11.5</td>
<td>-790</td>
<td>-700</td>
</tr>
<tr>
<td>5</td>
<td>154</td>
<td>On...</td>
<td>Rated</td>
<td>13.0</td>
<td>11.5</td>
<td>-790</td>
<td>-700</td>
</tr>
<tr>
<td>6</td>
<td>155</td>
<td>Off</td>
<td>Rated</td>
<td>13.0</td>
<td>11.5</td>
<td>-790</td>
<td>-700</td>
</tr>
<tr>
<td>7</td>
<td>156</td>
<td>On...</td>
<td>Rated</td>
<td>13.0</td>
<td>11.5</td>
<td>-790</td>
<td>-700</td>
</tr>
<tr>
<td>8</td>
<td>157</td>
<td>Off</td>
<td>Rated</td>
<td>13.0</td>
<td>11.5</td>
<td>-790</td>
<td>-700</td>
</tr>
</tbody>
</table>

* Reports on attempts to maximize loads (high side-slip amplitude).

**TABLE VI.—FISHTAIL MANEUVERS**

<table>
<thead>
<tr>
<th>Flight</th>
<th>Run</th>
<th>Speed (mph)</th>
<th>Power</th>
<th>$L$ (deg)</th>
<th>$\alpha_p$ (deg)</th>
<th>$\alpha_p$ (deg)</th>
<th>$\omega$ (deg/sec)</th>
</tr>
</thead>
<tbody>
<tr>
<td>1</td>
<td>200</td>
<td>On...</td>
<td>Rated</td>
<td>8.5</td>
<td>7.5</td>
<td>-1.30</td>
<td>-1.20</td>
</tr>
<tr>
<td>2</td>
<td>201</td>
<td>Off</td>
<td>Rated</td>
<td>8.2</td>
<td>7.5</td>
<td>-1.30</td>
<td>-1.20</td>
</tr>
<tr>
<td>3</td>
<td>202</td>
<td>On...</td>
<td>Rated</td>
<td>8.5</td>
<td>7.5</td>
<td>-1.30</td>
<td>-1.20</td>
</tr>
<tr>
<td>4</td>
<td>203</td>
<td>Off</td>
<td>Rated</td>
<td>8.2</td>
<td>7.5</td>
<td>-1.30</td>
<td>-1.20</td>
</tr>
</tbody>
</table>

* Reports on attempts to maximize loads (high side-slip amplitude).

**TABLE VI.—FISHTAIL MANEUVERS**

<table>
<thead>
<tr>
<th>Flight</th>
<th>Run</th>
<th>Speed (mph)</th>
<th>Power</th>
<th>$L$ (deg)</th>
<th>$\alpha_p$ (deg)</th>
<th>$\alpha_p$ (deg)</th>
<th>$\omega$ (deg/sec)</th>
</tr>
</thead>
<tbody>
<tr>
<td>1</td>
<td>300</td>
<td>On...</td>
<td>Rated</td>
<td>8.5</td>
<td>7.5</td>
<td>-1.30</td>
<td>-1.20</td>
</tr>
<tr>
<td>2</td>
<td>301</td>
<td>Off</td>
<td>Rated</td>
<td>8.2</td>
<td>7.5</td>
<td>-1.30</td>
<td>-1.20</td>
</tr>
<tr>
<td>3</td>
<td>302</td>
<td>On...</td>
<td>Rated</td>
<td>8.5</td>
<td>7.5</td>
<td>-1.30</td>
<td>-1.20</td>
</tr>
<tr>
<td>4</td>
<td>303</td>
<td>Off</td>
<td>Rated</td>
<td>8.2</td>
<td>7.5</td>
<td>-1.30</td>
<td>-1.20</td>
</tr>
</tbody>
</table>
Positive directions of axes and angles (forces and moments) are shown by arrows.

<table>
<thead>
<tr>
<th>Axis</th>
<th>Force (parallel to axis) symbol</th>
<th>Moment about axis</th>
<th>Angle</th>
<th>Velocities</th>
</tr>
</thead>
<tbody>
<tr>
<td>Designation</td>
<td>Symbol</td>
<td>Designation</td>
<td>Symbol</td>
<td>Positive direction</td>
</tr>
<tr>
<td>Longitudinal</td>
<td>X</td>
<td>Rolling</td>
<td>L</td>
<td>$Y\rightarrow Z$</td>
</tr>
<tr>
<td>Lateral</td>
<td>Y</td>
<td>Pitching</td>
<td>M</td>
<td>$Z\rightarrow x$</td>
</tr>
<tr>
<td>Normal</td>
<td>Z</td>
<td>Yawing</td>
<td>N</td>
<td>$X\rightarrow y$</td>
</tr>
</tbody>
</table>

Absolute coefficients of moment

\[
C_l = \frac{L}{\frac{q}{b}S} \quad C_m = \frac{M}{\frac{q}{c}S} \quad C_n = \frac{N}{\frac{q}{b}S}
\]

(rrolling) (pitching) (yawing)

Angle of set of control surface (relative to neutral position), $\delta$. (Indicate surface by proper subscript.)

4. PROPELLER SYMBOLS

- $D$: Diameter
- $p$: Geometric pitch
- $p/D$: Pitch ratio
- $V'$: Inflow velocity
- $V$: Slipstream velocity
- $T$: Thrust, absolute coefficient $C_T = \frac{T}{\frac{p}{n^2}D^3}$
- $Q$: Torque, absolute coefficient $C_Q = \frac{Q}{\frac{p}{n^2}D^3}$

\[
P \quad \text{Power, absolute coefficient } C_P = \frac{P}{\rho n^2 D^3}
\]

\[
C_s \quad \text{Speed-power coefficient } = \sqrt{\frac{s}{\rho n^2}}
\]

\[
\eta \quad \text{Efficiency}
\]

\[
n \quad \text{Revolutions per second, rps}
\]

\[
\Phi \quad \text{Effective helix angle } = \tan^{-1} \left( \frac{V}{2\pi n} \right)
\]

5. NUMERICAL RELATIONS

- 1 hp = 76.04 kg-m/s = 550 ft-lb/sec
- 1 metric horsepower = 0.9863 hp
- 1 mph = 0.4470 mps
- 1 mps = 2.2369 mph
- 1 lb = 0.4536 kg
- 1 kg = 2.2046 lb
- 1 mi = 1,609.35 m = 5,280 ft
- 1 m = 3.2808 ft